

The invention relates to rocket engineering and can be used as propulsion power plant of upper stages of rockets and acceleration blocks of rocket-carriers and space apparatuses. The liquid rocket engine with controlled thrust vector has a combustion chamber with nozzle, a turbo-pump system for fuel supply to the combustion chamber with turbine with exhaust collector of generator gas connected by means of blast tube to the circular collector of gas injection unit to the supersonic section of the nozzle, a circular injection slot connecting the circular collector to flow-through section of the nozzle, articulated joint of engine fixation to the body of the flight vehicle with drives of engine rotation for control of thrust vector in the pitching plane and course surface. The circular collector of the unit of exhaust gas injection and the circular injection slot are separated by partitions to two sections connected by blast tubes to exhaust windows of gas dispenser installed in the blast tube between the exhaust collector of the turbine and the collector for gas injection to the nozzle and connected to drive. In the circular slot profiled guide ribs are arranged, those are installed at angle to radial plane. In one section the ribs are made directed to one side, and in the other section – to the opposite side.