

A rotary turbojet blade comprises a plurality of blade sections extending along a line of the centers of gravity of said sections between a base and a tip, said blade presenting along a radial axis a bottom portion, an intermediate portion, and a top portion, said bottom portion presenting a longitudinal angle of inclination for a leading edge line, said intermediate portion presenting a backward longitudinal angle of inclination for said leading edge line, and said top portion presenting a backward longitudinal angle of inclination for said leading edge line and a tangential angle of inclination for said line of the centers of gravity in a direction opposite to the direction of rotation of said blade.