

A pitch control mechanism (controlling the angular position) for stator blades in a turbojet may be adjusted separately for two adjacent stages of blades controlled by a common actuator means 10. To achieve this, the synchronization bar 9 located between the mechanisms leading to the two stages is inclined, this inclination being determined by a pin 22 and groove 23 connection with the casing, while another pin 26 and groove 27 connection is arranged between the synchronization bar 9 and one of the mechanisms. Strongly non-linear displacement laws between the two mechanisms may thus be controlled depending on the shapes and directions of the grooves.