

The invention relates to cooling lining of afterburner chamber of aviation two-contour gas-turbine engine. Mixing lining (3) of flows of afterburning chamber is placed in the case (4). Between the case and the lining there is circular channel (5) intended for motion of secondary cold flow (F2). Front in direction of flow fuel injectors (6) are placed at inlet to diffuser (2) and flame stabilizers (7) are placed behind those. Casing of flow mixing (3) at section between the planes of front injectors and back flame stabilizers has doubled curvature and enlarges in direction of flow with possibility of retardation of primary gas flow (F1). Round the front in direction of flow section (10a) of mixing lining (3) there is circular bucket inlet (11) for intake of part of flow (F3) of air from cold flow (F2). Flow (F3) is supplied in tangent direction to primary flow (F1) by means of multiple channels (12) provided in the wall of the mixing lining (3) between the end inlet (11) and diffuser (2). The invention promotes cooling of mixing lining (3) for flows (F1) and (F2) at afterburning modes of operation of the engine.