

The invention relates to turbo-jet engine with sealing of cavity for air intake to cockpit restricted at one side with outer ring (6) of compressor, circular construction (7) connected to ring, and at the other side – with outer carter (12) of grid of diffuser, support construction connected to the outer carter and the outer ring (14) of carter of the engine fixed on circular construction (7) First gasket (50) is installed to the first slot (32) arranged round the front section (12a) of the outer carter (12) of grid of the diffuser, at that the plates of the first gasket rest on the back end of the first lug (71) arranged as one part with circular construction (7) Second gasket (60) is installed to the second slot (73) arranged under the circular construction (7), at that the plates of the second gasket rest on the front end of the second lug (72) arranged as one part with the circular construction, and the end of the third lug (76) provided on the front part (12a) of the outer carter pf the diffuser.